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A USER'S MANUAL FOR THE SEQUENCE ACCOUNTABLE FATIGUE ANALYSIS COMPUTER PROGRAM

J. M. POTTER R. A. NOBLE

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Structures Division

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trum of loads. Residual stress rel		
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analysis are lilustrated.		

FOREWORD

This program was prepared by J. M. Potter of the Solid Mechanics Branch and R. A. Noble of the Experimental Branch,

Structures Division, Air Force Flight Dynamics Laboratory. The work was conducted in-house under Project 1347 "Structural Testing of Flight Vehicles", Task 134704 "Structural Testing Criteria".

This report covers work accomplished over a time period of 1 October 1972 to 1 May 1973. The essence of the analysis was presented in AFFDL-TM-73-131-FBR in October 1973.

This manuscript was released by the authors in January 1974.

ABSTRACT

This report presents a detailed description of a computer program to calculate cumulative damage of notched structural members subjected to arbitrary spectra. The Sequence Accountable Fatigue Analysis computer program develops its sequence sensitivity by tracking residual stresses local to a notch throughout the spectrum of loads. Residual stress relaxation analysis is included to increase the generality of the results. An example spectrum and resulting cumulative damage analysis are illustrated.

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SYMBOLS

$^{\sigma}$ res	Residual stress
$\sigma_{ exttt{max}}$	Maximum local stress level
$\sigma_{ exttt{min}}$	Minimum local stress level
σys	Yield stress
$^{\sigma}\mathtt{res}_{ ext{EQ}}$	Equilibrium component of the residual stress
ε _t	Local strain, total
ε _e	Elastic component of total strain
ε p	Plastic component of total strain
ε _{f'}	Strain intercept at one reversal on a log ϵ_p -log life curve
С	Slope of the log $\varepsilon_{\rm p}$ -log life curve
S max	Maximum applied stress level
Smin	Minimum applied stress level
S _{mean}	Mean applied stress level, $(S_{max} + S_{min})/2$
K _t	Elastic stress concentration Factor
D	Damage
I	Integer describing the level number
N _{EP}	Equilibrium period, number of cycles for the local stresses to return approximately to the equilibrium conditions following an overload
C1	Residual stress relaxation constant
E1, E2	Relaxation function exponents
N	Number of cycles
E	Modulus of elasticity
N _f	Number of cycles of life at a given stress or strain cycling level

SECTION I

INTRODUCTION

Cumulative damage analyses based upon the local stress-strain behavior at a notch appear to be reasonably successful in anticipating trends in fatigue life behavior of notched specimens subjected to spectrum loading (1-6). The type of behavior that usually occurs is that peak tensile loads tend to increase the fatigue life and peak compressive loads tend to decrease the life of notched structures compared to structures experiencing load spectra not having those peaks (5,6). Local behavior analyses, such as those developed by Smith (7) and Neuber (8), help to explain this phenomenon as being a result of the tensile peak load creating a compressive residual stress at the notch and, conversely, the compressive peak creating a tensile residual stress. The change in life occurs because the residual stress state modifies the subsequent damage accumulation rates.

The Sequence Accountable Fatigue Analysis computer program was developed to incorporate the local stress-strain approach with a recent residual stress relaxation analysis (6) in order to improve the sequence sensitivity of cumulative damage analysis. This technical memorandum presents the details of the resultant computer program and an example of its use. The correlation of predictions made with this analysis to actual results of tests experiencing spectrum loading is presented by Potter (9), and Potter, Gallagher, and Stalnaker (10).

SECTION II

PROGRAM OUTLINE

The Sequence Accountable Fatigue Analysis traces the stress-strain behavior local to a notch throughout an applied load spectrum and calculates the damage based on the local experience. The computer program is divided generally into the four parts or modules outlined in Fig. 1.

The basic input data for the material, specimen geometry, fatigue behavior qualities and spectrum, are developed in Module I. The information required in Module I is discussed further in Section III. Module II takes the input information and determines the local stress-strain behavior. Module III references the Range Pair Counting Method Subroutine to cycle count the local stress spectrum. Module IV determines the damage in the local stress-strain spectrum.

The basic analyses used in Modules II, III and IV are presented below.

Module II - Local Stress-Strain Behavior - The analysis used during the determination of the local stress behavior during the spectrum of loading is a combination of analyses developed by Smith (7), Neuber (8) and Potter (6). Smith's simple analysis indicated that the residual stress could be approximated by assuming that the initial stress-strain behavior was elastic upon unloading following plastic flow. Thus, the residual stress could be calculated knowing the

maximum local stress and the maximum applied stress as in Eq. 1 and in Fig. 2.

$$\sigma_{\text{res}_i} = \sigma_{\text{max}_i} - \kappa_{\text{t}} S_{\text{max}_i} \tag{1}$$

The Sequence Accountable Fatigue Analysis computer program currently incorporates elastic-perfectly plastic stress-strain behavior. Therefore, σ_{max_i} is equal to the yield stress. For the cycles immediately following the peak stress, the residual stress determined in Eq. 1 modifies the elastic solution as shown in Eqs. 2 and 3 (provided that the following maximum applied stress is less than S_{max} and that there is no change in the residual stress due to a minimum applied stress causing reversed yielding).

$$\sigma_{\max_{i}} = \sigma_{\text{res}_{i-1}} + K_{t} S_{\max_{i}}$$
 (2)

$$\sigma_{\min_{i}} = \sigma_{\operatorname{res}_{i-1}} + K_{t} S_{\min_{i}}$$
 (3)

The analysis developed by Neuber $^{(8)}$ has been extended to cyclic loading by Wetzel $^{(2)}$ and Wetzel, Morrow and Topper $^{(3)}$ and used by many others $^{(1,4-6)}$ primarily to determine local stress-strain behavior. It is used in this program only to calculate plastic strains occurring when the residual stress undergoes a step change. The plastic strain calculation routine is accessed only when the σ_{\max_i} or σ_{\min_i} terms in Eqs. 2 and 3 exceed tensile or compressive yield stress levels, respectively. Figure 3 illustrates the calculation of the plastic strain.

The local stress-strain behavior, according to Wetzel (2) is related to the applied load by Eq. 4

$$\sigma \cdot \varepsilon = (K_{t \text{ max}})^2 / E$$
 (4)

The plastic strain can be found by subtracting the elastic component from the total strain.

$$\varepsilon_{\rm p} = \varepsilon_{\rm t} - \varepsilon_{\rm e} = (K_{\rm t} S_{\rm max})^2 / E \cdot \sigma_{\rm max} - \sigma_{\rm max} / E$$

Therefore, the plastic strain associated with S_{max_i} is given in Eq. 5.

$$\varepsilon_{p_i} = (K_t S_{max_i})^2 / E \sigma_{ys} - \sigma_{ys} / E$$
 (5)

If a residual stress existed prior to this plastic strain excursion, the plastic strain associated with that prior excursion is subtracted from Eq. 5 as shown in Eq. 6.

$$\varepsilon_{p_i} = (K_t S_{max_i})^2 / E\sigma_{ys} - (\sigma_{ys} - \sigma_{res_{i-1}})^2 / E\sigma_{ys}$$
 (6)

A similar calculation is made for plastic strains occurring during the minimum stress peak.

In the analysis developed by Potter (6) the residual stress cyclically relaxes toward zero or an equilibrium residual stress as shown in Fig. 4 according to Eq. 7.

$$\sigma_{\text{res}_{N=1},2,...} = (\sigma_{\text{res}_{N=1}} - \sigma_{\text{res}_{EQ}}) \exp(N/N_{EP_i} \ln(0.1))$$
 (7)

The $N_{
m EP}$ term, the Equilibrium Period, is dependent upon the applied stress and the Residual Stress Relaxation Constant.

$$N_{EP_{i}} = (C1/\left|K_{t}S_{max_{i}}\right|^{E1} \cdot \left|K_{t}S_{mean_{i}}\right|^{E2})$$
 (8)

The Residual Stress Relaxation Constant, C1, has not yet been experimentally defined but should be a constant for a material.

Module III - Cycle Counting Method

After the local stress and plastic strain behavior is calculated, the local stress spectrum is Range Pair Counted using a computer program developed by Tischler. (11)

Module IV - Damage Calculation

Damage is calculated separately for the plastic strain excursions and the elastic stress spectrum. The damage is determined from the conventional D = $\sum_{N=1}^{\infty} \frac{1}{N}$ calculation. Damage from each of the plastic strain cycles is determined from the Coffin-Manson (12) form

$$D_i = 1./N_{f_i} = 1./(\epsilon_{p_i}/\epsilon_{f_i})^{1/c}$$

Damage from the elastic stress cycles is determined in a similar manner. The maximum and minimum local stress levels are sequentially compared to unnotched S-N data in a Modified Goodman Diagram format. Damage is summed, and failure of the coupon is defined as the event occurring when the summed damage equals unity.

SECTION III

INPUT DATA REQUIREMENTS

In general, each spectrum analyzed will require slightly different programming in order to get the load history into a usable format for the core program. The basic program requires a certain family of information before any analytical predictions can be made. Appendix I contains a program listing for the Sequence Accountable Fatigue Analysis. The subroutine CORE which accesses the subroutines having to do with RPCM, the Range Pair Counting Method, contains the basic analysis. Subroutine SAL reads the data input and then references subroutine CORE. The subroutine SAL shown is one in which a block of cycles is repeated with optional cycles. A list of the input data cards and the resulting analysis is given in Appendix II.

The specific data requirements are given below.

- 1. Stress-Strain Behavior The stress-strain behavior is presumed to be elastic-perfectly plastic with the tensile yield stress being equal to the compressive yield stress. The yield stress value used is an average of the monotonic behavior generally being above the 0.2% yield value and below the engineering ultimate strength.
- 2. Residual Stress Relaxation The residual stress relaxation behavior of Eq. 7 and 8 is characterized by Cl, the Residual Stress Relaxation Constant and El and E2, the relaxation equation exponents. The Residual Stress Relaxation Constant, Cl, has not yet been adequately determined. It should be a material property if the relaxation function

is correct and must be assumed. A reasonably accurate estimate of the Residual Stress Relaxation Constant for aluminum material falls in the range of 5-20 x 10⁶ (cycles) (Ksi)². Further experimentation on the part of the analyst should develop a Cl usable for his set of conditions until actual measurement of residual stress relaxation behavior defines the relaxation function and constants. At present El and E2 are considered to be equal to 1.0.

- 3. Specimen Geometry The elastic K_t value (if available) is entered into the analysis. If that value is not available then an estimate from some other method may be used. In certain cases, a value may be determined from a constant amplitude fatigue test of a similar structure by fitting several values of K_t to the analysis and determining the best correlation as is done with the K_f solution. Once a stress concentration factor, K_t , is determined for a specimen, that value is not changed from test-to-test of the same coupon donfiguration.
- 4. <u>Load Multiplier</u> Different spectra are presented for analysis in different manners. Some data are presented in percent of maximum stress, others in terms of nominal stress, and others in terms of bending moment. The value of the load multiplier defines the nominal stress history.
- 5. Cumulative Damage Analysis The damage from the range-paired elastic stress spectrum is determined by calculating a simple $\frac{n}{N}$ value for each level and accumulating the total. The N value is determined from unnotched coupon S-N data in the Modified Goodman Diagram format.

The program requires the input of four second order equations describing the maximum and minimum stress levels at lives of 10^4 , 10^5 , 10^6 and 10^7 cycles. The coefficients of the equations are derived by least square fitting the S-N data presented in the form of Eq. 9.

$$S_{max} = A(I) S_{min}^{2} + B(I) S_{min} + C(I)$$
 (9)

The A, B, and C coefficients for several typical materials are presented in Appendix IV. The S-N data shown was derived from various sources but usually from the MIL-HDBK-5A $^{(13)}$. The C coefficients correspond to the maximum stress level at zero to maximum applied stress conditions on the unnotched coupons.

The damage from the plastic strain cycles is determined using the Coffin-Manson relation to calculate the N $_{\rm f_1}$ value. The conventional plastic strain intercept at one reversal and the $\epsilon_{\rm p}$ - life slope values are used in the analysis. Specific measured values from the literature are used when available and typical values when they are not available.

6. Analysis or Test Spectrum - The last information needed is the order and magnitude of application of the spectrum used in the test.

SECTION IV

OUTPUT OPTIONS

The Computer Program prints the following output in the process of the analysis.

- 1. Maximum and minimum applied stress and local stress response through the spectrum. Also printed out is the residual stress, equilibrium stress, applied cycles, and the equilibrium period.
- 2. The elastic local stress history as input into the Range Pair Subroutine and the resulting Range Paired spectrum.
- 3. The plastic strain occurrence during the spectrum and the damage associated with each strain reversal.
 - 4. The accumulated damage associated with the plastic strains.
- 5. The Range Paired elastic stress spectrum and the damage associated with each level.
- 6. The accumulated damage associated with the current block of loading including the plastic strain damage and the total damage since the initiation of cycling.

At the option of the analyst, he can print out all the above items or only two. The IPRINT value controls what data is printed.

- If IPRINT = 1, all six items are printed for each flight or block.
- If IPRINT = 2, all items except 2. above are printed.
- If IPRINT = 3, only items 4. and 6. above are printed.

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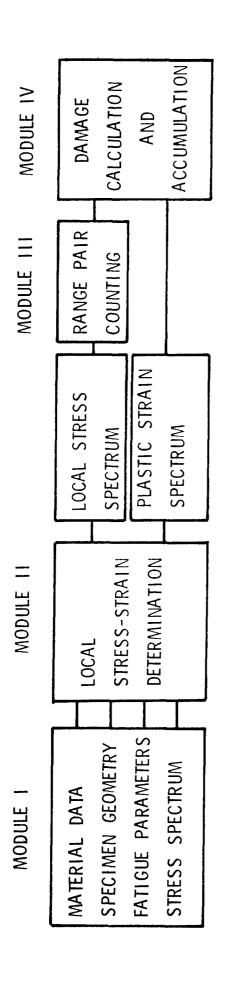


FIGURE 1. PROCEDURE USED IN THE SEQUENCE ACCOUNTABLE FATIGUE ANALYSIS

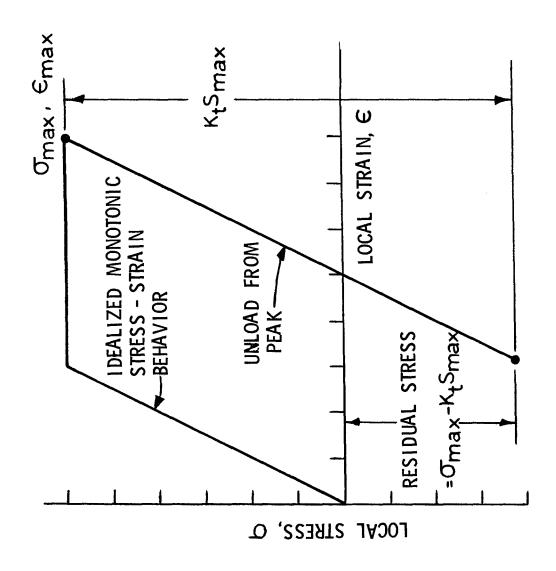


FIGURE 2. METHOD OF DETERMINING THE RESIDUAL STRESS FOLLOWING A PEAK LOAD

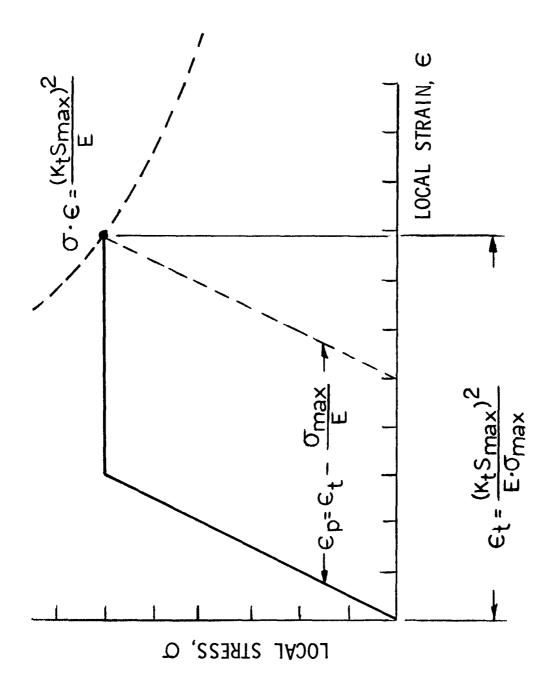


FIGURE 3. METHOD OF DETERMINING PLASTIC STRAIN LEVELS

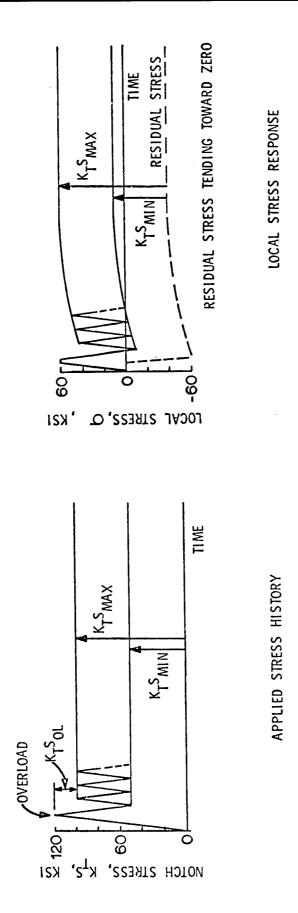


FIGURE 4. LOCAL STRESS RESPONSE FOR APPLIED CONSTANT AMPLITUDE LOADING WITH RESIDUAL STRESS RELAXATION

(CONSTANTS TO BE USED IN CALCULA-)

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3DC 6600 FTN V3.0-367A OPT=1	IF(JA+JB) 230,250,240 RES(I)=TYS-ASMAX GO TO 290	(I)==1YS-ASHIN TO 290	JA) 260,260,270	(I)=RES(I=1)	No. 1 (1)	MAX(IXAIN) =RES(I)+ASHAX	MIN(IRAIN)=RES(I)+ASHIN	YC(IRAIN) = ENN(J)	ASMAX.LE. TYS) 60 TO 410		TO 440	ASMIN.GE17S) GO TO 430		10 440	ਦੱਲ = 0 •	=RES(I)-EQRES	NOTIONAL BY ANALTON BUILDING		AX=ABS(ASMAX)	ABHIN=ABS (ASHIN)	MANIE DOS (ASSERTA)	(ABMAX.LT.1.) ABMAX=1.	(ABMEAN.LT.1.) ABMEAN=0.5	ABMIN-ABMAX) 444,444,442	=ABMIN	GO TO 446	TABMAX	P=C1/(ABM**E1*ABMEAN**E2)		WRITE(6,350) STMAX(J), STMIN(J), SIGMAX(IRAIN), SIGMIN(IRAIN),	ENN(J), ENEP, J, IR	MAI(6(F/.2,1X), F6.2,1X,E15.8,5X,I6,4X,I6)	#ON T - NOO		CALCOLAS O KONTOCAL DIKODO KELDXAJION	* + NI + OI = NI	T. 25.5 (0.15)	10 360	10 560	00	1000.*ENN(J).LT.ENEP) GO TO 560	(J).LE.10.) GO TO 560	D I SQ I O I SQ I SQ	A62=10	IKAIN+IRAIN+1		IF(OUMMY=ENEP) 470,460,460	11 = UUMIV 2 •	AG=NFLAG+1	
CORE		240 RE	250 IF		2.0	1S 062	SI	S.	I.F.	400 EQ							ى د	ى د		AB	8₽	H	H		442 AB		444 444		H.T.	3 ;			100) (. .		4	09	00	360 IF	G		380 28	2	¥ ;		450 IF		2	
SUBROUTINE		9			,	. 59					7.0					75				8.0					85				(عر بر				ď					100					102				440	110	

CORE 11.5					CURE 135 CORE 136 CORE 137 CORE 138		CORE 145 CORE 146 CORE 147 CORE 148 CORE 149	CORE 151 CORE 152 CORE 153 CORE 155 CORE 155 CORE 156 CORE 156	GORE 150 CORE 151 CORE 152 CORE 164 CORE 155
CYCINT=DUMMY/10. DO 500 K=1,10 DECK=E: OAT(K)	EN(K)=CYCINT*DECK IF(K,EQ.1) GO TO 490 EX(K)=EXP(-2.303*EN(K-1)/ENEP)+EXP(-2.303*EN(K)/ENEP) GO TO 500	<pre>D EX(K)=1.+EXP(-2.303*EN(K)/ENEP) CONTINUE IF(NFLAG.EQ.0) GO TO 530 D NFLAG2=NFLAG+10 DO 520 K=11,NFLAG2</pre>	EX(K)=E.*OUMHY EX(K)=E.XP(-2.303*EN(K-1)/ENEP)+EXP(-2.303*EN(K)/ENEP) OUMHY=2.*DUMHY CONTINUE DO 559 K=1,NFLAG2	SIGHIN(IRAIN)-ASDANATEURESTOLFTEXIK)/Z. SIGHIN(IRAIN)-SIGMAX(IRAIN)-ASMAX+ASHIN RNCYC(IRAIN)=ENKY) IF(K.EQ.1) GO TO 540 RNCYC(IRAIN)-RNCYC(IRAIN)-EN(K-1)	MRITE (6,950) SIGMAX (IRAIN), SIGMIN (FRAIN), RNCYC (FRAIN), IRAIN FORMAT (16H RELAXATION ,2(F7.2,1X),16X, F6.2,31X,16)	IRAIN=IRAIN+1 CONTINUE CONTINUE RES(I)=EQRES+DIF*EXP(-2,303*ENN(J)/ENEP) CONTINUE RES(1)=RES(J)	IN=IRAIN-1 ++++++++++++++++++++++++++++++++++++	CALL SUBROUTINE TO RANGE PAIR COUNT SPECTRUM IF (IRPCM.GT.1)GO TO 591 GO TO 592 CONTINUE	GONIINUE CONIINUE ************************************
0.24	084	490 500 510	91 87 84 80 9 90	10 10 10 10 10 10 10 10 10 10 10 10 10 1	550 551	559 560 570 569	00000	591	0000 0000
		120	125	130	135	140	14 75 60 60	155	160

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·		1000 1000 11000	168
	10 552	CORE	170
	100111 (6,593)	CORE	171
ń	SULTING FATIGO	CORE	172
	*** TECT TOWN THE TEMPLEY TOWN THE STRAIN TOWN TOWN OR HIN TOWN OF	CORE	173
ŭ		CORE	174
175 C		CORE	175
ပ	CALCULATE DAMAGE FROM PLASTIC STRAIN CYCLES	9 14 00 0	1/0
O		CORF	178
		CORE	179
	JKL=1,NLEVEL	CORE	130
101		CORE	181
	1 (PLS KA (JKL) 532,531,533	CORE	182
2 K	Z NIT-I. DICTOALIVITATES CTDALIVI	CORE	183
		CORE	194
185		בטאנו ממקור	185
		ה ה ה	186
		CORE	187
	70 524	ב ה ה ה	138
		CORE	189
190 53	S WRITE (6.199) JRI. PI STRACIRI), DAN	1 CO 2 C	190
19	9 FORMAT (10X-14-12X-F10-5-15X-34MTN-15X-F14-6)	3 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6	191
	60 10 531	יין אַר כּיַר קיני	761
53	7 WRITE(6,219)JKL,PLSTRA(JKL),DAM	CORE	7.7
2	FORMAT(10X,I4,12X,F10.5,15X,3HMAX,15X,E14.6)	CORE	125
195 5.	CONTINUE	CORE	136
i	WRITE(6,541) SUMDEL	CORE	197
ŗ	ROM PLASTIC STRAINS=,E15,8)	CORE	198
	3.56	CORE	199
200 13	FORM IT AND TO SEE THE STORY OF SEE AND SEE AN	CORE	200
•	VAIDED HOVEN AND THE TOTAL	COXE	201
53	6 CONTINUE	יו אמני ווי	202
O		מאט ר מינים ר	503
	LEAST SOUARE FITTED N-N DATA	1,00	707
205		CORE	707
O		CORE	202
į	DO 600 JKL=1, KPMAX	CORE	208
ň		CORE	209
210	SIGMIN(JKL).LT.1.6*TYS)60 TO 310	CORE	210
		CORE	211
31	0 IF (SIGHAKUKL) GF. TIYS) GO TO 220	CORE	212
	7 CYCLES=10.**9.	1 400	21.5
•	60 10 340	ORE	215
215 32	411X 0	CORE	216
		OORE	217
	X:N)=x:N)+016HIN(OXC)++X+R(N)+NIGHIN(OXC)+C(N)-NIGHXX(OXC)	CORE	218
33	HILLOCU D	CORE	219
		ti no	

SUBRO	SUBROUTINE CORE		CDC 6600 FIN V3.0-367A OPT=1	04/26/74	10.40.58.	PAGE	ľ
	334	ABR7=ABS(R(7))		CORE	222		
		A8R4=A3S(R(4))		CORE	223		
		IF(ABR7.LE.A.BR4.) GO TO 336		CORE	224		
	335	EXP0=4.+R(4)/(R(4)-R(5))		CORE	225		
225		GO TO 339		CORE	526		
	336	EXP0=7.+R(7)/(R(6)-R(7))		CORE	227		
				CORE	228		
	338	SUMR=R(4)+R(5)+R(6)+R(7)		CORE	229		
)			CORE	230		
230	:	-		CORE	231		
		٠		CORE	232		
\$ 1 may 2 m				CORE	233		
		SUMR2N=4,*R(4)**2+5,*R(5)**2+6,*R(6)**2+7,*R(7)**2	2**1	CORE	234		
		DEL1=4. *SUMR2*SUMR4-4. *SUMR3**2		CORE	235		
235		DEL2=SUNR*SUMR2*SUMR3-SUMR4*SUMR**2		CORE	236		
				CORE	237		
		DO1=22.*SUMR2*SUMR4-22.*SUMR3**2		CORE	238		
		DO2=SUMR2*SUMR3*SUMRN-SUMR*SUMR4*SUMRN		CORE	239		
		DO3=SUMR*SUMR3*SUMR2N-SUMR2N*SUMR2**2		CORE	2+0		:
240		EXP0=(001+D02+D03) / (DEL1+DEL2+DEL3)		CORE	241		
	339	CYCLES=10.**EXPO		CORE	242		
;	:	IF (EXPO.LE.4.) CYCL SS=10. **4.		CORE	243		!
_	340	ENNCYC=RNCYC (JKL) / CYCLES		CORE	544		
		SUMNC=SUMNC+ENNCYC		CORE	542		
242		SUMDEL = SUMDEL + ENNCYC		CORE	546		
		:11		CORE	24.7		
		HRITE(6,399) SIGMAX(JKL),SIGMIN(JKL),RNCYG(JKL),CYCLES,ENNCYG), CYCLES, ENNCYC	CORE	248		
	665	FORMAT (16X, 2 (F7. 2, 1X), 16X, F6. 0, 17X, 2 (E15.8,1X))	2	CORE	549		
	009	CONTINUE		CORE	250		
250		WRITE(6,593) SUMDEL		CORE	251		
	593	FORMAT (/69X, 21H DAMAGE PER THIS SET=, £15.8)		CORE	252		
		SUMNC		CORE	253		
	575	FORMAT (/69x, 18H TOTAL ENN/CYC =, £15.8)		CORE	554		
				CORE	255		

256	255 259 259	252 262 263 263 263	265 265 257 257	268 272 272 273	275	23 2 2 3 3 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5	8 4 4 6 5 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6	\ \ \ \ \ \ \ \ \ \ \ \ \ \ \ \ \ \ \	222 222 320 320 322 332 332 332 332 332	0.48.95.00.00.00.00.00.00.00.00.00.00.00.00.00
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	METHOD TO GENER	NS.)	DIMENSION NPKS + KK DING INPUT CYCL	+ + + X X X X X X X X X X X X X X X X X X X	NPKS + + + + + + + + + + + + + + + + + + +	. 66	4,KIND		INPUT LOAD SPE INIMUM,13X, I = 1,NPKS)	KS AND VALLEYS
RPCH(NPKS)	LOYS THE RANGE PAIR CYCLE COUNTING H FROM A GIVEN LOAD SPECTRUM	PROGRAM ARRAYS (INFORMATION NEEDED TO CHANGE DIMENSIONS)	DEFINITION PEAKS OF THE INPUT LOAD SPECTRUM THE NUMBER OF ADJITIONAL CYCLES (EXCLUDING WHICH THE PROCRAM WILL GENERALE	3 3 6 6 6 6 6	K COUNTERS OF THE COCKES OF THE UNSORTED ANALYSIS SPECTRUM STEP NUMBERS OF 31 FMENTS OF THE	RTED ANALYZIS SPECTRUM DES OF NSTEP(J) SUCH THAT R 1.0 AND VALUES OF NSTEP(J SIGMAX(J-1) = SIGMAX(J) A IIN(J-1) = SIGMIN(J)	FRNCYC(200),KPMAX,IPRINT L/SIGMAX(200),SIGMIN(200) Z/NSTEP(200),KS,KMAX,KMIN,K31 Z/NSTEP(450),INDEX(450),IND1,IND2,IND3,IND4,KIND	ND	NUT NUT A/31 K/) P(I)	THE LOAD SPECTRUM
SUBROUTINE	THIS PROGRAM EMPI ANALYSIS SPECTRU		ARRAY NAME Sigmax KK :	SIGHIN RNCYC NSTEP RES INDEX	N N E C Y C	ISAVE	COMMON/MSAL/RNCY/ COMMON/MDEC1/SIG- COMMON/MDEC2/NST COMMON/MDECZ/NST COMMON/MDECR/RES	000 666	MRITE(6,20) NPKS 20 FORMAT(1H0,60HTH 1CTRUH = ,15//) MRITE(6,22) 22 FORMAT(6,25) WRITE(6,25) WRITE(6,25) Z5 FORMAT(29X,15,10)	SORT THROUG COUNTER K I
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CORE	CORE	CORE	CORE	CORE	CORE	CORE	CORE	CORE	CORE	CORE	C026	CORE	CORE	CORE	CORE	C02E	0.00 1.00 1.00 1.00 1.00 1.00 1.00 1.00	CORE	CORE	CORE	CORE	CORE		CORE	CORE	CORE	CORE	CORE	503 110 110 110 110 110 110 110 110 110 1	1 to 0.0	C03E	CORE	CORE	CORE	1200	יין אַנ מיין	CORE	CORE	CORE	CORE	300	3
		0 60 70 100		,RNCYC(I),NSTEP(I))				00	9 K H 190MAX) 101	RS OF THOSE PE	R K IS LESS THAN 1.0//(1717))				0 110		1+1)						SPECTRUM DATA - COMBINE STEPS WITH IDENTICAL	CONSECUTIVEL			SIGMAX(I-1)) GO TO 300	60 10		٠		000				244		I+1)	I+1)		17)	
# II	0 = 100 = 0	\sim	X1 = SIGMAX(I) X2 = SIGMIN(I)	CYCGEN(X1,X2	1 = 0 + 1	100 CONTINUE	NPKSN = NPKS - JHAX		WKILE(5)23) (ISAVE(K)) IF(IPRINT,GE,2)GO TO 1	23 FORMAT (1H0,93HSTEP NUMBERS	1SPECTRUM WHOSE COUNTER		VE(U) -	7	IF (I .EQ. NPKN) GO TO	. ·	(TETT) VERDIO - (TI) VERDIO (TETT) - (TETT) VERDIO (TETT)	-	RNCYC(II) = RNCY	115 CONTINUE	ALC CONTINCE		SORT THROUGH THE L	C AND VALLEYS WHICH OCCUR		00 300 I = 2, NPKSN	IF (SIGMAX(I) .NE. SIG		DANACATAN - BANKA (1-4)	1	_		ا 1	$311 \ J = 1, JMA$		TE CI LED NOKN) CO TO	316 II = I.NPKN	SMAX(II)		NSTEP(II) = NSTEP(II+1)	18	1000 010
		9			65			Ç T	.			75				6	3			ı,	85 2			6	÷				45				100				105				110	

SUBROUTINE	ITINE RPCM COC 6600 FIN V3.0-367A OPT=1	47792740	10.40.58.	PAGE	(
	311 CONTINUE NPKSN = NPKSN - JMAS C	CORE	356		
115	Z " "	C C C C C C C C C C C C C C C C C C C	3350 375 371 372		
120	A 4 1 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2	CORE CORE CORE CORE	373 375 376 376		
125	IF (KB .NE. 0) 60 T0 5 X1 = SIGMAX(I) X2 = SIGMIN(I) IND1 = NSTEP(I) IND2 = IND1	20000 20000 4 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2	3 3 3 3 3 3 3 3 3 3 3 3 3 3 3 3 3 3 3		
130	I = I + 1 KB = 1 GO TO 1 S X3 = SIGMAX(I) X4 = SIGMIN(I)	000000 000000 000000000000000000000000	7 4 10 00 P- 0 0 80 80 80 1 80 80 80 80 1 80 80 80 80	:	
135	IND3 = NSTEP(I) IND4 = IND3 KMIN = 1 KMAX = 0	0000 0008 008 008 008	3 3 3 3 3 3 3 3 3 3 3 3 3 3 3 3 3 3 3		
140	* X " " O "	0000 0000 0000 0000 0000	1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1		
145	CYCNO = AINT(RNGYC(I)+0.5) CALL DECIDE(X1,X2,X3,X4,KEY,I,GYCNO,KCYGEN) 1000 GO TO (10,10,30),KGYGEN 10 KB = 1 10 KB = 1 10 KB = 1	CORE CORE CORE	10000 10000 110000 110000		
150	IF (KMIN ,NE, 1) GO TO 36 IF (I , LE, NPKSN) GO TO 5 RES(LR+1) = X1 RES(LR+2) = X2 INDEX(LR+1) = IND1	C C C C C C C C C C C C C C C C C C C	2 4 6 5 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6		
155		CORE CORE CORE CORE	. 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4		
160	IF (I • LE• NPKSN) GO TO 31 RES(LR+1) = X1 RES(LR+2) = X2 RES(LR+3) = X3 INDEX(LR+1) = IND1 INDEX(LR+2) = IND2	CORE CORE CORE	0 4 0 0 0 t		
165	INDEX(LR+3) = IND3	CORE	420		

	SUBROUTINE		30C 6600 FTN	V3.0-367A OPT=1	04/26/74	-3*	PAGE	. 	
KAIN = 0	m	LRMAX = LR + 3 G0 T0 2000 X + = SIGHAX(I) IND4 = NSTEP(I) KMAX = 1			000RE 000RE 000RE 000RE	00000			
0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0		KMIN = 0 K31 = 1 IF (RNCYC(I) .GT. 1.0) GO TO 4			CORE CORE CORE	N W N			:
GOTO GOTO GOTO GOTO GOTO GOTO GOTO GOTO		GO TO 6 40 KEY = 1 KIND = 0			002E	NWW		:	
#### #################################		0 X			00000 00000 000000) M M I			· i
Second		T NOTE II NOTE			CORE	~ ~ ~			
No.		K31 = 0 60 T0 32	; ;		CORE	~ ~ ~			
CONTRACT		36 X3 = SIGMIN(I)			CORE	n t m			
V		KAILN II II KAILN II			CORE	. i. t		:	
X1 = SIGHAX(I) X2 = SIGHAX(I) X3 = SIGHAX(I) X4 = SIGHAX(I) X4 = SIGHAX(I) IND1 = INSTEMIC) IND2 = IND1 IND3 = IND1 IND4 = IND3 IND4 = IND4 IND4 = IND5 IND5 = IND		60 10 12 KeY = 1 TF (KB .NF. 0) GO TO 44			00 K	* * :	•	:	
X = SIGHANII) X = SIGHANII) X = SIGHANII) IND1 = NNSTEPLI) IND2 = NNSTEPLI) IND3 = NNSTEPLI) IND3 = NNSTEPLI) IND4 = IND1 IND3 = NNSTEPLI) IND4 = IND1 IND3 = NNSTEPLI) IND4 = IND1 IND5 = NNSTEPLI) IND4 = IND1 IND5 = NNSTEPLI) IND6 = NNSTEPLI) IND7 = NNSTEPLI) IND8 = IND1 IND8 = IND3 IND8 = IND1 IND8 = IND		-			CORE	t 4			
X4 = SIGHIN(I) IND1 = IND1 IND2 = IND1 IND3 = IND1 IND3 = IND1 IND3 = IND1 IND3 = IND1 IND4 = IND3 IND3 = IND1 IND4 = IND3 IND3 = IND1 IND5 = IND1 IND5 = IND1 IND6 = IND1 IND7 = IND3 IND7 = IND7 IND8 = IND8 = IND7 IND8 = IND8 = IND8		XZ = SIGMIN(I) X3 = SIGMAX(I)			CORE	t t		•	!
INDZ = INDI INDZ = INDZ INDZ =		X4 = SIGMIN(I) IND1 = NSTEP(I)			CORE	J 10			
Number N		IND2 = IND1			CORE	-in 1		:	
KHAN = 1 KHAN = 0 KHAN = 1 KHAN = 1 KHAN = 0 KHAN = 1 KHA		TONI =			CORE	ומוח	:	;	
K31 = 0		KAIN = 1 Kaax = 0			CORE	וט וט			
RNCYC(I) = RNCYC(I) - 1.0 GO TO 402 MINCYC(I) = RNCYC(I) - 2.0 GO TO 402 MINCYC(I) = RNCYC(I) - 2.0 GO TO 415 GORE 452 CORE 453 CORE 453 CORE 455 IND4 = IND3 KMIN = 1 KMIN = 1 KMIN = 1 KMIN = 1 CORE 465 CORE 465 CORE 465 KMIN = 1 CORE 467 CORE 467 CORE 467 CORE 467 CORE 467 CORE 467 CORE 470 CORE 470 CORE 470 CORE 477		.LE. 2.03 GO TO			CORE	10 10	•		1
1 RNCYC(I) = RNCYC(I) - 2.0 2 KIND = 0 6 O TO 415 10 X = SIGMAX(I) 11 X = SIGMAX(I) 12 X = SIGMAX(I) 13 X = SIGMAX(I) 14 X = SIGMAX(I) 15 X = SIGMAX(I) 16 X = SIGMAX(I) 17 X = SIGMAX(I) 18 X = SIGMAX		- 1.0			COR.	1 10 1			•
D2 KIND = 0 CORE		RNCYC(I) = RNCYC(I) - 2.			CORE	U 'U			:
10 X3 = SIGMAX(I) X4 = SIGMIN(I) X4 = SIGMIN(I) 1 ND3 = NSTEP(I) 1 ND4 = IND3 KMIN = 1 KMAX = 0 KXIN = 1 RHCYC(I) = RNCYC(I) - 1.0 CORE 470 CORE 470 CORE 471 CORE 471 CORE 472 CORE 473 CORE 474 CORE 475 CORE 475 CORE 476 CORE 477		KINU GO T			CORE	יט ינ			
X4 = SIGMIN(I) IND3 = NSTEP(I) IND4 = IND3 IND4 = IND3 KMIN = 1 KMIN = 1 KNIN = 1 CORE		. II			CORE	, 10			:
IND4 = IND3 KMIN = 1 CORE		X4 = SIGNIN(I) IND3 = NSIED(I)			CORE	ດທ			
KMIN = 1 CORE 467 KMAX = 0 CORE 458 KA1 = 0 KIND = 1 RHCYC(I) = RNCYC(I) - 1.0 CORE 470 CORE 471 CORE 472 CALL DECIDE(X1,X2,X3,X4,KEY,I,CYCNO,KCYGEN) CORE 477 CALL DECIDE(X1,X2,X3,X4,KEY,I,CYCNO,KCYGEN) CORE 477 CORE 477 CORE 477		IND # FONI			CORE	'0		-	· ·
K31 = 0 KIND = 1 KIND = 1 KORE 470 CORE 471 KHCYC(I) = RNCYC(I) - 1.0 CORE 471 CORE 472 CORE 472 CORE 472 CORE 472 CORE 473 CALL DECIDE(X1,X2,X3,X4,KEY,I,CYCNO,KCYGEN) CORE 473 CORE 475 CORE 475		11 11			CORE	യം			
KIND = 1 COKE +70 KHCYC(I) = RNCYC(I) - 1.0 KB = 0 CYCNO = AINT(RNCYC(I)+0.5) CALL DECIDE(X1,X2,X3,X4,KEY,I,CYCNO,KCYGEN) COTE +74 COTE +75 COTE +75 COTE +75		K31 = 0	4		CORE			1	:
KB = 0 CYCNO = AINT(RNCYC(I)+0.5) CYCNO = AINT(RNCYC(I)+0.5) CALL DECIDE(X1,X2,X3,X4,KEY,I,CYCNO,KCYGEN) GO TO 1000		= RNCYC(I) -			CORE	~ ~			
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		60 10 1000		e e e e e e e e e e e e e e e e e e e	CORE	. ~	1		

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SUBROUTINE RPCM CDC 6600 FIN V3.0-367A OPT=1	2000 LMAX = L IF (LRMAX .LT. 4) GO TO 5000 IF (NCYNO .EQ. 0) GO TO 5000	C RANGE PAIR COUNT OF RESIDUE SPECTRUMS C	NRES = NRES + 1	5 0 0 0	G COUNT THE LAST RESIDUE SPECTRUM - RANGE PAIR COUNTING WILL YIELD C ADDITIONAL CYCLES C	KK = 0 RESMAX = RES(1) RESMIN = RES(1) TMAX = 4	IMIN = 1 DO 500 I = 2, LRMAX IF (RES(I) .LT. RESMAX) GO TO 490	RESMAX = RES(I) IMAX = I GO TO 500 490 IF (RES(I) • GI • RESMIN) GO TO 500	RESMIN = RES(I) IMIN = I CONTINUE	CALL CYCRES(RESMAX,RESMIN,1.0,INDEX(IMAX))	KK = KK + 1 IMAX = J GO TO 510 550 J = IMIN + 2 TF (1) GT + DMAX) GO TO 676	S(RES(J-1), RES	1X = L RT THE ANALYSIS SPECTRUM TO PRODUCE THE RANGE PAIR COUNTED = 0	UU DUS JJ = 1,NPKS KC = 0 00 600 I = 1,LMAX IF (NNSTEP(I) •NE. JJ) GO TO 600 KP = KP + 1 KC = KC + 1 NSTEP(KP) = KP	1
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30C 6600 FTN V3.0-367A 0PT=1 04/26/74 10.40.58.	SIGHAX (KP) = CYCLE (I,1)	SIGMIN(KP) = CYCLE(I,2)	RNCYC(KP) = RNEGYC(I)	:	X(KP-1)) G0	IF (SIGMIN(KP) .NG. SIGMIN(KP-1)) GO TO 600		RNCYC(KP) = RNCYC(KP) + 1.0	CONTINUE	605 CONTINUE	KPHAX # KP	IF (IPRINT.6E.2)60 TO 104	WRITE(6,2010)	2010 FORMAT(1H1,48X,33HRANGE PAIR CYCLE COUNTED SPECTRUM//)	HRITE (6.22)	HKITE (6, 25) (NSTEP (I), SIGMAX (I), SIGMIN (I), RNCYC(I), I H 1, KPMAX)	102 FORMAT(5X*3F10*2)	CONTINUE	END
SUBROUTINE RPCM							595		909	609				2010			102	104	
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CYCGEN CYCGEN COC 6600 FIN V3.0-367A OPT=1	SUBROUTINE CYGGEN(Y1,Y2, CYCPF,NSTEPP) COMMON/MCYG/CYCLE(200,2),RNECYC(200),NNSTEP(200) COMMON/MCGDE/L,LIND C THIS SUBROUTINE GENERATES CYCLES FOR THE ANALYSIS SPECTRUM FROM DA		CYCLE(L,1) = Y1 CYCLE(L,2) = Y2 RNECYC(L) = CYCPF NNSTEP(L) = NSTEPP IF (L,F0,1) GO 100	IF (CYCLE(L-1,1) NE. CYCLE(L,1)) 60 TO 100 IF (CYCLE(L-1,2) NE. CYCLE(L,2)) 60 TO 100 10 L = L - 1 RNECYC(L) = RNECYC(L) + 1.0	LIND = 1 100 RETURN END
SUBROUTINE CYCGEN	w		o	ι ν	0

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	7 KEY, IN (201 AX, KM) 50), IN	UM SAI	X1) G0	(3)	X1) G0		: •		-
	X3, X4, SIGH	S WHETHER SPECTRUM	T. EP (O, NSTEP(I)	.;		SPECTRUM		TO 110 500), KCYGEN) GO TO 1151 RETURN 0
	X1,X2,X3,X4,KEY,I, X(200),SIGHIN(200) (200),LR,KMAX,KHIN 50),INDEX(450),IND 200,2),RNECYC(200)	CIDE	100 m 4 Z		T0 1	TURN	OUE.		TO 1
	SUBROUTINE DECIDE(X COMMON/MDEC1/SIGMAX COMMON/MDEC2/NSIEP(COMMON/MDECR/RES(45 COMMON/MCYG/CYCLE(2 COMMON/MCYG/CYCLE(2 COMMON/MCYGDE/L,LIND	SUBROUTINE DE THE ADJUSTED THE ADJUSTED THE DE DE DE GO	x2) 60 x1) 60 x4 .0R x3 60 (x3,x2,	(2,X3,	0) 60 X4 •OR	. RE	HÉ RESI I IND1		ET (KEY .NE. D) GO RETURN GO TO (1150,1200,1 IF (CYCNO .GT. 1.0 CYCNO = CYCNO - 1. GO TO 1152
	INE DE MDEC1/ MDEC2/ MDECR/ MCYG/C	BROUT;	61.)	YCGEN () 1 4 03 • NE • INO4	= 1 • NE• • GT•)	11. IND4 IND4 CYCNO	F d×n	TINDS INDS IND¢	(1150,12 (1150,12 (CNO 6T- NO LE- EYCNO
	BROUT HHON/ HHON/ HHON/	THIS SUR FROM THE KFIRST =	בר גגה בר גגה	CALL CY X1 = X1 X2 = X4 X2 = X4 IF (IND	KCYGEN = 1 IF (KEY •NE• RETURN IF (XZ •GI• GO TO 150	S KEN X X	ADD X1 TO ADD X1 TO LR = LR + RES(LR) = INDEX(LR) X1 = X2	XXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXX	TF (KEY * RETURN 60 TO (11 ECYCNO CYCNO CY
DECIDE	W 00 0 0 0	- F X H	10 IF 11 IF 150 IF CAL	151 CALL CYGGEN(X2,X3, 1, 152 X1 = X1 X2 = X4 IF (INO3 .NE. IND2) L: IND2 = IND4	75 17 210 17 60	Z00 X1 X2 X2 X2 X2 X2 X3 X3 X3 X3 X3 X3 X3 X3 X3 X3 X3 X3 X3	SOO AD SIN IN I		1150 IF 1150 IF 1153 IF
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SUBROUTINE	DECIDE	COC 6600 FIN V3.0-367A OPT=1	4/192/14	10.40.58. PA	PAGE	2
	1151 IF (LIND .EQ. 1) GO TO 1153 IF (IND3 .NE. IND4) GO TO 1153 RNECYC(L) = RNECYC(L) + CYCNO - 2	2.0	CORE CORE CORE	626 627 628		
0 9	1152 IF (KMAX NE. 1) GO TO 111 X3 = SIGMIN(I)		CORE CORE CORE	629 530 631		
65	INUS = NSIEP(I) IF (CYGNO .GT.0.0) GO TO 112 KMIN = 1 KMAX = 0		008E	\$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$		
	KCYGEN = 3 RETURN 1200 IF(CYCNO .LE. 0.0) RETURN CYCNO = CYCNO - 1.0		0000 0000 0000 0000 0000 0000 0000 0000 0000	1		į
0 2	X3 = SIGMAX(I) X4 = SIGMIN(I) KFIRST = 1 GO TO 113 414 X3 - CIGMAX(I)		C C C C C C C C C C C C C C C C C C C	35 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5		
22	III XX = SIGNIN(I) X4 = SIGNIN(I) IF (KFIRST .NE. 0) GO TO 113 GYCNO = GYCNO = 1.0 KFIRST = 1		CORE CORE CORE	7027 2000 2000		:
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89 22	1500 IF (KRAX NE. 0) 60 TO 1510 IF (CYCNO - LE. 0.0) RETURN CYCNO - 1.0 112 X4 = SIGMAX (I) INO4 = NSTEP(I) KMAX = 1		0008 008 008 008 008	6 6 5 5 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6		
06 6 96	KHIN = 0 GO TO 11 INDE X = SIGMIN(I) INDE = NSTEP(I) KMAX = 0 KMIN = 1 GO TO 10 END		0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	7 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0		

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10.+0.58.	6569 670 671 672	4 C G G G G G G G G G G G G G G G G G G	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	\$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$	699 691 692 993 994	6995 6995 6997 6996 0	701 702 703 704	705 707 708 708	7117	7 + 6
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CDC 6600 FTN V3.0-367A O	ICLRMAX, NCYNO) ND (450), INDEX(450), IND1, IND2, IND3, IND4, KIND FCIDES WHETHER OR NOT THE ELEMENTS OF THE ITHE RANGE PAIR COUNTING CONDITIONS			TO 100 • X3 • GT• X1) GO TO 500 TO 151 1.0,IND3)	1.0,IND2)	TO 300 K) GO TO 315	3 .LT. X1) GO TO 500			
DECRES	SUBROUTINE DECRES(LRM COMMON/MCGDE/L,LIND COMMON/MDECR/RES(450) THIS SUBROUTINE DECIDION SPECTRUM SATISFY THE	K = 0 NCYNO = 3 X1 = RES(1) X2 = RES(2) X3 = RES(3)	X4 = RES(4) IND1 = INDEX(1) IND2 = INDEX(2) IND3 = INDEX(3) IND4 = INDEX(4)	J = 4 IF (XZ .6T. X1) GO IF (XZ .LT. X4 .0R IF (XZ .6T. X3) GO CALL GYDRES(X3,XZ,	GALL CYCRES(X2,X3, NCYNO = NCYNO + 1 X1 = X1 X2 = X4	" E EE #	IND4 = J+1	G0 T0 150 0 K = K + 1 RES(K) = X1 INDEX(K) = IND1 J = J + 1 IF (J 61, RMAX) G0 T0	1 = X2 2 = X3 3 = X4 4 = RES(J) NO1 = TND2	1007 + 1007
SUBROUTINE DE	000 0	3 .		150	151		100	005		
8	: 	10	15	20	:	30	32	04	45	

INDEX(K) = IMD1 INDEX(K) = IMD2 CORE TETUR AFTUR RES(K) = X1 RES(K) = X1 RES(K) = X1 RES(K) = IMD1 INDEX(K) = IMD1	SUBROUTINE	DECRES	COC 6600 FIN V3.0-367A OPT=1	04/26/74 10:40.58.	10.40.58.	PAGE	8
CORE CORE CORE CORE CORE CORE CORE CORE		INDEX(K) = INDI		CORE	723		
CORE CORE CORE CORE CORE CORE CORE CORE		INDEX(K+1) = IND2		CORE	724		
CORE CORE CORE CORE CORE CORE CORE CORE		LKMAX II X + 1		CORE	725		
((J+1) ((J+1) ((ORE CORE		RETURN		CORE	726		
(0xe coxe coxe coxe coxe coxe coxe coxe co		315 K = K + 1		CORE	727		
CORE CORE CORE CORE CORE CORE CORE CORE		RES(K) = X1		CORE	728		
CORE CORE CORE CORE CORE CORE CORE CORE		RES(K+1) = X2		CORE	729		
100 CORE ND2 ND2 ND2 ODE CORE CORE CORE CORE CORE CORE CORE COR		RES(K+2) = RES(J+1)		CORE	730		
NOEX (J+1) CORE CORE CORE CORE CORE CORE CORE COR		INDEX(K) = IND1		CORE	731		
40Ex (J+1) CORE		INDEX(K+1) = IND2		CORE	732		
CORE CORE CORE CORE ND3 ND4 CORE CORE CORE CORE CORE		INDEX(K+2) = INDEX(J+1)		CORE	733		
CORE CORE CORE CORE CORE CORE CORE CORE		LRMAX = K + 2		COR	734		
CORE CORE CORE NO3 ND4 OORE CORE CORE CORE		RETURN		CORE	735		
CORE CORE CORE CORE CORE CORE CORE CORE		330 K = K + 1		CORE	736		
CORE CORE CORE CORE CORE CORE CORE CORE		RES(K) = X2		CORE	737		
CORE CORE CORE CORE CORE CORE CORE CORE		RES(K+1) = X3		CORF	738		
NO3 CORE CORE CORE CORE CORE CORE CORE		RES(K+2) = X4		CORF	739		
EX(K+1) = IND3 EX(K+2) = IND4 AX = K + 2 CORE CORE CORE CORE CORE CORE CORE CORE		INDEX(K) = IND2		CORF	7.50		
EX(K+2) = IND4 AX = K + 2 CORE URN CORE		INDEX(K+1) = IND3		CORE	761		
AX = K + 2 CORE URN CORE CORE		INDEX(K+2) = IND¢		1800	742		
URN CORE CORE CORE		LRMAX = K + 2		100	21:2		
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SUBROUTINE	CYCRES	CDC 6688 FIN V3.0-367A OPT=1 84/26/74 18.48.58.	04/26/74	10.40.58.	PAGE	•
	SUBROUTINE CYCRES(Y1, Y2, CYCPF, NSTEPP)	EPP)	CORE	942		
	COMMON/MCYG/CYCLE(200,2), RNECYC(200), NNSTEP (200)	CORE	2+2		
•	COMMON/MCGDE/L, LIND		CORE	248		
ပ			CORE	642		
2	THIS SUBROUTINE GENERATES CYCLES FO	R THE ANALYSIS SPECTRUM FROM (DA CORE	750		
ပ	SUPPLIED BY SUBROUTINE DECRES		CORE	751		
O			CORE	752		
	1 + 1		CORE	753		
	CYCLE(L,1) = Y1		CORE	754		
10	CYCLE(L,2) = Y2		CORE	755		
	RNECYC(L) = CYCPF		CORE	756		
	NNSTEP(L) # NSTEPP		CORE	757		
	RETURN		CORE	758		
	CNL		FECO	759		

APPENDIX II

SAMPLE PROBLEM WITH INPUT DATA LISTING

1 DATA DECKS ARE TO BE PROCESSED.

SPECTRUM SUBJECTED TO THE RANGE-PAIR COUNTING TECHNIQUE SEQUENCE ACCOUNTABLE FATIGUE EVALUATION

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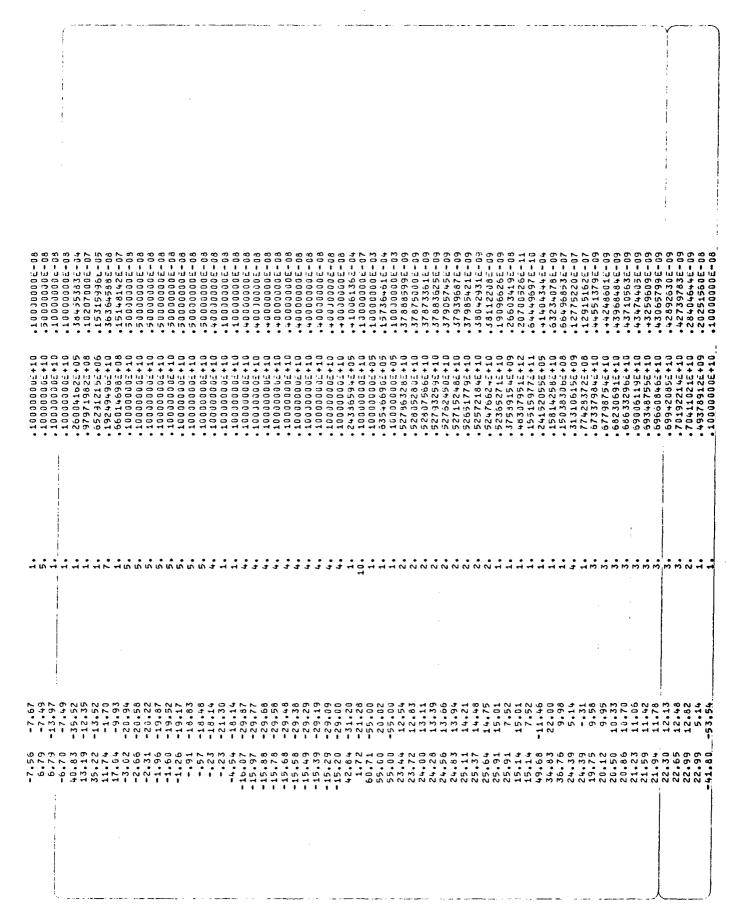
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APPENDIX III

LIST OF COMPUTER PROGRAM SYMBOLS AND DEFINITIONS

A Coefficient of the x^2 term in the equation of a line on a constant life fatigue diagram where minimum stress is x and maximum stress is y. ($R = Ax^2 + Bx + C - y$)

AA An assigned value of +1. or -1.

AAA A stress used in the calculation of plastic strain.

ABDIF The absolute value of DIF.

ABM The absolute value of ASMAX or of ASMIN, as assigned.

ABMAX The absolute value of ASMAX.

ABMEAN The absolute value of ASMEAN.

ABMIN The absolute value of ASMIN.

ABR4 The absolute value of R(4).

ABR7 The absolute value of R(7).

ABS The name of a routine calling for the absolute value of a quantity.

AKT Stress concentration factor, K

ASMAX The product (AKT) (STMAX)

ASMEAN The quantity (ASMAX + ASMIN)/2

ASMIN The product (AKT) (STMIN)

AVSGMN Average value of SIGMIN over an interval.

AVSGMX Average value of SIGMAX over an interval.

B Coefficient of the x term. (See A.)

BBB A stress used in the calculation of plastic strain.

C The constant. (See A.)

COFMAN Inverse of the Coffin-Manson slope.

CYCINT The number of cycles in an interval.

CYCLES The calculated number of cycles expected to be indicated on a constant life fatigue diagram for the applied combination of maximum and minimum stress.

Cl The Residual Stress Relaxation Constant (See ENEP.)

DAM Damage.

DECK Decimal or real value of integer K after conversion.

DEL2 A portion of a least-squares-method solution.

DIF The difference between residual stress and equilibrium residual stress. (RES(I) - EQRES)

DO2 A portion of a least-squares-method solution.

DUMMY A variable used in the calculation of the number of cycles to be considered as an interval for relaxation determination.

ELMOD The elastic modulus.

EN The number of cycles from the beginning of the relaxation process to the end of the current interval.

ENEP The number of cycles required for overload residual stress effect to return to within one-tenth of its original difference from equilibrium conditions.

$$(N_{ep} = C1/(ABM)^{E1}(ABMEAN)^{E2})$$

ENN The number of applied cycles at a load level.

ENNCYC The ratio of the number of applied cycles to the number

of cycles to failure. (ENN/CYCLES)

EPSD LCF strain intercept.

EQRES Equilibrium residual stress.

EX An exponential function depicting the relaxation of

residual stress.

EXP The name of a routine calling for the exponential value

of a quantity.

EXPO An exponent. The power of 10 which indicates the number

of cycles to failure.

E1 Residual stress Relaxation Exponents.

FLOAT The name of a routine calling for integer-to-real conversion.

I A variable subscript.

IBLOCK The identifying number of a block the blocks being

numbered consecutively from 1 to NBLOCK.

IFIX The name of a routine calling for real-to-integer conversion.

IN The number of steps input to the range pair counting

subroutine.

IPRINT Value controlling the WRITE statements.

IRAIN A counter.

IRPCM Value controlling entry into the range pair counting

subroutine.

ISTEP The identifying step number, the steps being numbered

from 1 to NLEVEL.

ITYPE The identifying type number, the types being numbered

from 1 to NTYPE.

J A variable subscript.

JA Value of +1 or 0, as assigned for branch determination.

JB Value of -1 or 0, as assigned for branch determination.

JJ An index variable.

JJJ An index variable.

JKL An index variable.

K An index variable.

KK An index variable.

KPMAX The number of steps output from the range pair counting

subroutine.

L An index variable.

LMN An index variable.

M An index variable.

N An index variable with values of N=4-7 indicating the

power of 10, and thus identifying a particular life cycle

curve.

NBLOCK The total number of times to execute a block of loads.

NDECK The number of data decks to be run sequentially.

NFLAG An integer used as a counter.

NFLAG2 An integer used as a counter.

NLEVEL The total number of steps, or levels, of loads in a block.

NN A subscripted variable used to indicate which types of

loads are experienced in which blocks.

NTYPE The total number of different types.

PLSTRA Plastic strain.

R Residue term in damage calculation.

RES Residual stress.

RNCYC The number of cycles for a level after exitting the range

pair counting subroutine.

SIGMAX Maximum stress.

SIGMIN Minimum stress.

STMAX Maximum applied stress.

STMIN Minimum applied stress.

SUMDEL Summation of damage for a flight.

SUMENN Accumulated total of applied cycles. (Summation of ENN).

SUMNC Accumulated cycle ratio. (Summation of ENN/CYCLES).

SUMR Summation of R(N), N=4,7.

SUMRN Summation of NR(N), N=4,7.

SUMR2 Summation of $R(N)^2$, N=4,7.

SUMR2N Summation of $NR(N)^2$, N=4,7.

SUMR3 Summation of $R(N)^3$, N=4,7.

SUMR4 Summation of $R(N)^4$, N=4,7.

TITLE1, TITLE2 Identification of the source of the SN data.

TLL Tensile load limit.

TM1, TM2 Material type.

TTYS One-fifth of tensile yield stress.

TYS Tensile yield stress.

T1,T2,T3,T4,T5,T6,T7,T8 Test identifying information.

X Variable equivalent to SIGMIN.

Y Variable equivalent to SIGMAX.

APPENDIX IV

FATIGUE LIFE INPUT DATA FOR SEVERAL MATERIALS

MATERIAL	YIELD STRESS	STRAIN INTERCEPT	INVERSE OF SLOPE	LIFE,	S-N LI A(I)	FE COEFF B(I)	C(I)
2024-T4	58.	0.4 ②	-1.8362	4	0020	.2091	62.63
				5	0032	.4366	51.4
				6 7	0035 0042	.6207 .7003	42.2 36.1
				,	.0042	• 7003	_
2219-T851	55.	0.4 ②	-1.8362	4	0022	.2204	55.84
				5	0018	.3320	48.3
				6	0015	.4628	39.7
		_	_	7	0024	.6420	31.7
7075 - T6	72.	0.4 ①	-1.836	4	0020	.2801	71.7 ³
	-				002 2	.5154	56.3
				5 6	0014	.6141	44.6
				7	0013	.6838	38.1
DOG 100	105	0.545	-1.493	,	0.0	0106	98.3
RQC-100	125.	0.54	-1.493	4	0.0	.2136	98.3 88.5
				5 6	0.0 0.0	.2927 .3669	79.1
				7	0.0	.4376	70.3
				,	0.0	•4370	
Man-Ten	55.	1.11	-1.6675	4	0.0	.2257	63.5
				5	0.0	.3520	53.1
				6	0.0	.4669	43.7
				7	0.0	.5678	35.4
4340 Stee1	160.	0.42	-1,836 ²	t.	0002	.2567	162.43
4340 Steet	160.	0.4	-1,030	4 5	0007	.5248	126.9
				6	0007	.5557	113.5
				7	0005	.5557	108.5
		<u></u>		•			
Ti-6-4	158.	0.42	-1.836	4	0009	.2368	154.2
				5	0006	. 4640	110.3
				6	0000	.4650	88.9
				7	.0001	.4752	84.2

¹ Data from Endo, T., and Morrow, J., NAEC-ASL-1105, Naval Air Engineering Center, Philadelphia, PA, June 1966.

Data not available - Source 1 data considered typical.

Derived from Metallic Materials and Elements for Aerospace Vehicles Structures, MIL-HDBK-5A, Dept. of Defense, Washington, D.C., February 1966. Information supplied from Rockwell International.

Information supplied by Society of Automotive Engineers Cumulative Damage Division, Courtesy of Mr. H.R. Jaekel.